

Amendments To The Claims:

Please amend the claims as shown.

1 – 7 (canceled)

8. (new) An axial flow gas turbine engine arranged about a central axis, comprising:
a compressor section;
a combustion chamber;
a turbine section having a plurality of guide vane rings and rotor blade rings arranged in axial succession in a hot-gas duct that contains a hot gas flow;
a cooling air flow for cooling the guide vane rings and the rotor blade rings, where the pressure of the cooling air flow decreases in the direction of the hot gas flow; and
a sealing element arranged between a guide vane ring and a directly adjacent rotor blade ring which seals the different pressure levels associated with the respective adjacent rings and extends as a single piece around at least a quarter of a circle concentric with the central axis of the engine.
9. (new) The gas turbine as claimed in claim 8, wherein the sealing element extends half of the circle.
10. (new) The gas turbine as claimed in claim 9, wherein the sealing element is formed as an annular metal sheet with a surface extending in the radial direction and having an outer and an inner edge.
11. (new) The gas turbine as claimed in claim 10, wherein the sealing element inner edge is arranged in grooves within a guide vane that is located in the side furthest from the hot-gas duct and the outer edge is arranged in a carrier groove.
12. (new) The gas turbine as claimed in claim 11, wherein the sealing element is clamped to the groove side wall using a screw which presses the sealing element onto the opposite platform groove side wall and carrier groove side wall.

13. (new) The gas turbine as claimed in claim 12, wherein the guide vanes each have an axial fixed point at which they are fixed against axial displacement in the guide vane carrier by a hooked formation, with the sealing element being arranged in the region of the axial fixed points.

14. (new) The gas turbine as claimed in claim 10, wherein the guide vanes each have an axial fixed point at which they are fixed against axial displacement in the guide vane carrier by a hooked formation, with the sealing element being arranged opposite the region of the axial fixed points.

15. (new) An axial flow gas turbine engine sealing system, comprising:
a guide vane or carrier ring having a groove; and
a sealing element arranged between a guide vane ring and a directly adjacent rotor blade ring which seals the different pressure levels associated with the respective adjacent rings and extends as a single piece around at least a quarter of a circle concentric with the central axis of the engine.

16. (new) A semi-annular turbine blade and vane stage seal, comprising:
an inner seal edge arranged in a guide vane groove that is located distal a hot gas duct;
an outer seal edge arranged in a vane carrier groove;
a metallic seal sheet having an arc length between 90° and 180°;
wherein the stage seal isolates different cooling flow pressure levels associated with an adjacent stage.